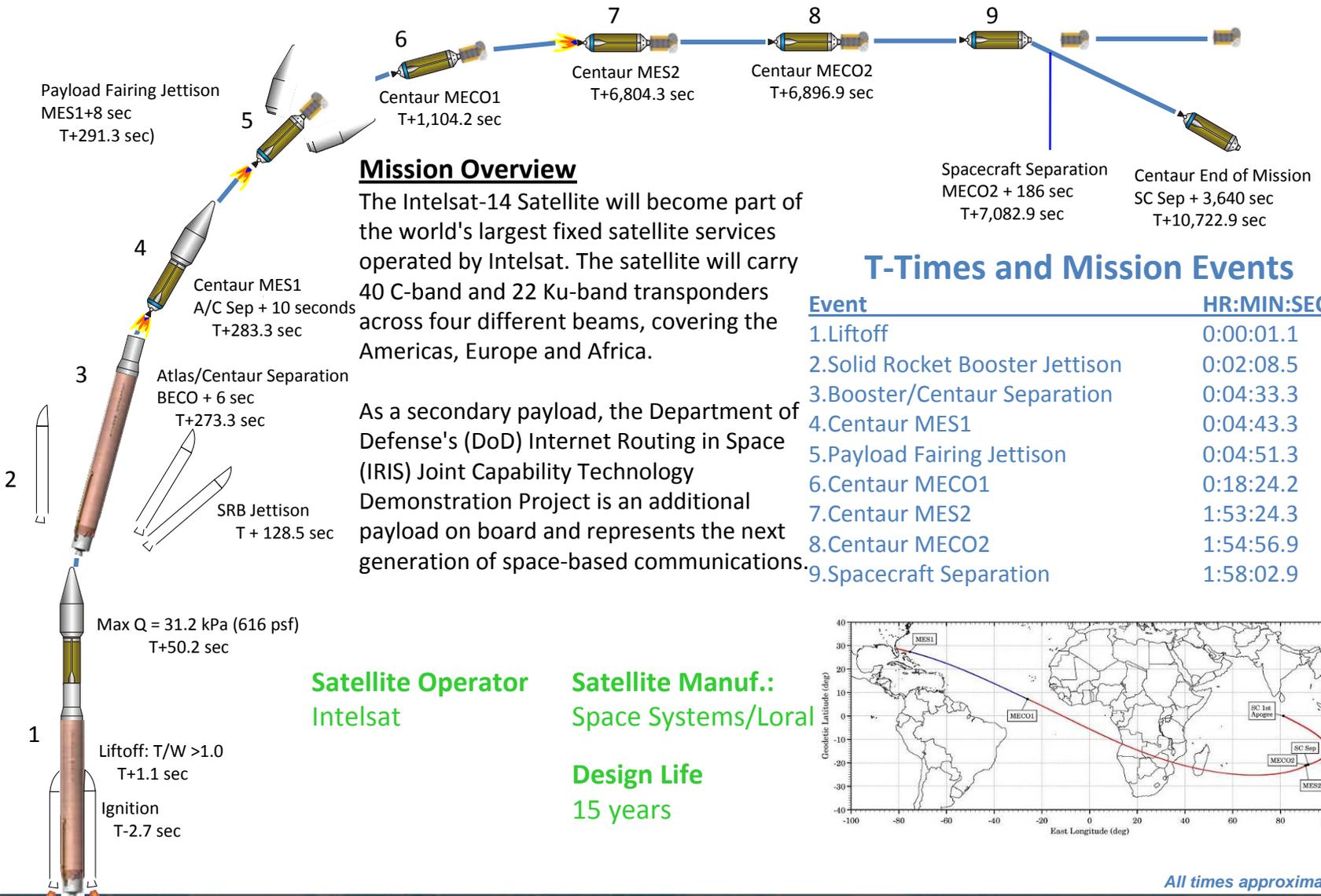


Mission Overview



Mission Overview

The Intelsat-14 Satellite will become part of the world's largest fixed satellite services operated by Intelsat. The satellite will carry 40 C-band and 22 Ku-band transponders across four different beams, covering the Americas, Europe and Africa.

As a secondary payload, the Department of Defense's (DoD) Internet Routing in Space (IRIS) Joint Capability Technology Demonstration Project is an additional payload on board and represents the next generation of space-based communications.

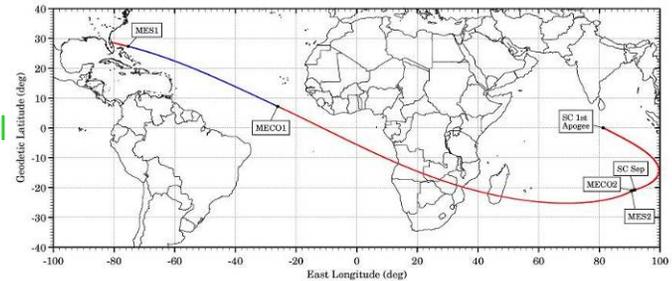
T-Times and Mission Events

Event	HR:MIN:SEC
1.Liftoff	0:00:01.1
2.Solid Rocket Booster Jettison	0:02:08.5
3.Booster/Centaur Separation	0:04:33.3
4.Centaur MES1	0:04:43.3
5.Payload Fairing Jettison	0:04:51.3
6.Centaur MECO1	0:18:24.2
7.Centaur MES2	1:53:24.3
8.Centaur MECO2	1:54:56.9
9.Spacecraft Separation	1:58:02.9

Satellite Operator
Intelsat

Satellite Manuf.:
Space Systems/Loral

Design Life
15 years



All times approximate



Mission Overview

INTELSAT 14

1. The Intelsat-14 mission will be flown from Launch Complex (LC-41) at Cape Canaveral, Florida on an Atlas V launch vehicle (431 configuration, tail number AV-024) with three solid rocket boosters (SRB) and a single engine Centaur stage. The Intelsat-14 satellite will be encapsulated in a 4-meter diameter extra extended payload fairing (XEPPF) and integrated to the Centaur using a 47-inch diameter payload adapter (PLA), a low shock payload separation system, and electrical and separation system harnesses.
2. The mission will commence at T-5 seconds when the aft plate ejects from the launch vehicle, followed by the Atlas RD-180 engine ignition at T-2.7 seconds. SRB ignition takes place at T+0.8 seconds.
3. Liftoff occurs at T+1.1 seconds. Shortly after the vehicle clears the launch pad, it performs a programmed pitch/yaw/roll maneuver. Max Q occurs at T+50.2 seconds into flight.
4. Vehicle telemetry and data are gathered and relayed through various local and down-range tracking stations. The tracking and Data Relay Satellite System (TDRSS) also serves to downlink vehicle telemetry throughout the mission.
5. Centaur separates 6 seconds after Booster Engine Cutoff (BECO) at T+273.3 seconds. Centaur main engine Start (MES1) occurs 10 seconds after the separation event at T+283.3 seconds. Payload fairing jettison occurs at T+291.3 seconds, 8 seconds after MES1. At T+1,104.2 seconds, Centaur main engine shutdown occurs (T+1,104.2 seconds) and the Centaur has achieved the first stage of its parking orbit.
6. After a 95 minute coast phase, Centaur reorients itself for engine re-ignition. Main Engine Start 2 (MES2) begins at T+6,804.3 seconds. Main Engine cutoff (MECO 2) occurs 95.6 seconds later at T+6,899.9 seconds.
7. Following another short coast phase, the Centaur will begin to reorient its attitude in preparation for space craft separation at T+7,082.9 seconds. Centaur end of mission occurs at T+10,722.9 seconds.

